

ATTY DOCKET NO.
13DV-13738-04

DOC. ID
DV13738-4B

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In the Claims:

Please amend the Claims as follows:

32. (currently amended) An aircraft gas turbine engine assembly comprising:

axially spaced apart fan and inter-turbine frames rotatably supporting co-axial high and low pressure rotors,

each of said frames comprising a first structural ring and a second structural ring, said second structural ring disposed co-axially with and radially spaced inwardly of said first structural ring about a centerline axis, and a plurality of circumferentially spaced apart struts extending radially between said first and second structural rings,

said inter-turbine frame axially located between high and low pressure turbines of said high and low pressure rotors respectively,

said high pressure turbine located forward of said low pressure turbine,

said low pressure rotor includes a low pressure shaft which is at least in part rotatably disposed co-axially with and radially inwardly of said high pressure rotor,

aft sump member having an aft central bore fixedly joined to said inter-turbine turbine frame,

said low pressure rotor supported by an aftwardmost ~~second~~ bearing mounted in said aft central bore of said aft sump member,

said high pressure rotor partly supported by a differential bearing mounted in an annular recess that is located radially inwardly of said aftwardmost ~~second~~ bearing,

~~said annular recess extends axially aftward into a radially enlarged portion of said aft end, and~~

a frame connecting means for connecting said engine to an

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aircraft located on said first structural ring.

33. (currently amended) An assembly as claimed in claim 32 wherein said low pressure turbine has an axial center of gravity and is connected to said low pressure shaft at or near said aftwardmost second turbine frame bearing.

34. (original) An assembly as claimed in claim 33 wherein said low pressure turbine includes a turbine disk assembly comprising axially adjacent rotor disks interconnected by structural disk forward and aft spacer arms, respectively, and said turbine disk assembly is connected to said low pressure shaft at or near said axial center of gravity.

35. (currently amended) An assembly as claimed in claim 34 further comprising a conical shaft extension drivingly connecting said turbine disk assembly to said second low pressure shaft and said conical shaft extension connected to said turbine disk assembly at or near said axial center of gravity.

36. (currently amended) An assembly as claimed in claim 35 wherein said turbine disk assembly further comprises said disks having pluralities of hubs connected to rims by webs extending radially outwardly from said hubs and each of said rotor disks supports includes a row of blades supported in said disk rim.